MOD Format 756B
(Established Jul 24)

Aircraft Weighing Report

Single-Point Weighing

Serial No.	Load Reading	Lat Corr. Fig	Lat Corr. Load	Cal. Corr.	Actual Load	Weight of Tackle at Rotor Head	Symbol	Aircraft Weight as Weighed
							W	
Distance from Main Reaction point to:		Nose ..... Front ..... Jacking Point		Tail ..... Rear ..... Lifting Point			**L**	FWD/AFT (Del. As req'd)
Weight added to bring A/C Longitudinal Datum Line Level* Or Tail Reaction*							**Wt1**	*Delete As Req'd
Distance from Main Reaction Point to Aircraft * Datum Point:							**d**	Fwd/Aft (Delete As req'd)
Distance of C of G from Main Reaction Point as Weighed = $\frac{(Wt1) \times (L)}{(W)}$							**a**	Fwd/Aft (Delete As req'd)
Distance of C of G from _____ * Datum as Weighed = (d) + or - (a)							**x**	Fwd/Aft (Delete As req'd)
Aircraft Moment as Weighed = (W) x (x) (**+** If C of G **Aft** of Datum, **-** If C of G **Fwd** of Datum)							**M**	Fwd/Aft (Delete As req'd)
Basic Weight of Aircraft = W (**+** Weight of Items **Deficient** to Basic State) (**-** Weight of Items **Surplus** to Basic State)							**BW**	
Aircraft Moment in Basic Condition About _____ * Datum = M (**+** Resultant Moment of Items **Deficient** to Basic State) (**-** Resultant Moment of Items **Surplus** to Basic State)							**BM**	
Distance of C of G from _____ * Datum in Basic Weigh Condition = $\frac{(BM)}{(BW)}$							**X**	
							%MAC	

* Insert Correct Datum from Aircraft Manual

Note: 1. Resultant Moment of Surpluses and Deficiences must be applied separately.

2. Where 2 separate stacks of weights are used to balance Aircraft, the Moment of each set of weights must be calculated separately.