MOD Format 756(Apache)
(Established Jul 24)

Aircraft Weighing Report

Multi-Point Platform Corrected Method

Latitude Correction Figure	

Weighing Position	Platform Serial No.	Indicated Load	Indicated Zero Error	Latitude Correction Load	Calibration Correction	Actual Load
Main Port						
Main Stbd						
Tail						
				First Weigh Total	**W**	
Main Port						
Main Stbd						
Tail						
				Second Weigh Total	**W**	
Main Port						
Main Stbd						
Tail						
				Third Weigh Total	**W**	

Mean Main Pt U/C Weight	lbs	**Wmr L**	
Mean Main Stbd U/C Weight	lbs	**Wmr R**	
Mean Tail Weight	lbs	**Wtr**	
Mean of 3 recorded Aircraft Total weights	lbs	**W_T**	

Longitudinal Calculations			
Horizontal distance from CrossTube c/l to Mainwheel c/l	ins	**Mr**	
Horizontal distance from CrossTube c/l to Tail Gear c/l	ins	**KTr**	
As Weighed C of G from Cross Tube (Wmr L*Mr)+(Wmr R*Mr)+(Wtr*KTr)/(W_T)		**Xa**	
Aircraft out of level angle as a decimal	degrees	**Ø**	
Vertical C of G from Main Cross Tube	ins	**z**	
Corrected C of G - (Xa/CosØ) - Z x TanØ)	ins	**x2**	
Distance from WRD to C of G Datum	ins	**B**	
Converted C of G from C of G Datum = x2 + B	ins	**x**	
As Weighed moment = (x) x (W_T)	lbs ins	**m**	
Basic Weight = (W_T) plus deficiences, minus surpluses	lbs	**BW**	
Basic Moment = m plus deficiences, minus surpluses -115-1034 (Tail and Main U/C Correction)	lbs ins	**BM**	
Basic Centre of Gravity from Aircraft datum point	ins	**X**	

Lateral Calculations			
Centre Port Main Wheel to Aircraft Centreline:	ins	**L3p**	
Centre Stbd Main Wheel to Aircraft Centreline:	ins	**L3s**	
Mean Main Pt U/C Weight	lbs	**Wtm L**	
Mean Main Stbd U/C Weight	lbs	**Wtm R**	
Aircraft Lateral as weighed C of G = (WtmL x L3p)+(WtmR x L3s)+(Wtr x 0)/(W_T)	ins	**y**	
Aircraft Moment as weighed = (W_T) x (y) (+ if C of G Stbd of Datum, - if C of G Pt of Datum)		**m(y)**	
Basic Weight of Aircraft = W_T plus deficiences, minus surpluses		**BW**	
Basic Moment = m(y) plus deficiences, minus surpluses		**BM(Y)**	
Distance of C of G from: A/C * Datum in Basic Weight Condition = $\frac{(BM(Y))}{(BW)}$		**Y**	

% Mean Aerodynamic Chord (%MAC) =	**% MAC**	