Aircraft Weighing Report

MOD Format 756F

Multi Point Platform Incidence Corrected Method (Harrier)

(Revised Jun 01)

Aircraft Type	Mk	Serial Number

Last Recorded basic figures from MOD Form 751 / WRAM			
Weight (Lbs)	**Moment (Lbs/ins)**	**C of G (ins)**	**% MAC**

SNCO IC Aircraft	
I Certify that the Surplus and Deficiency form (MOD Form 756D) has been completed and where necessary signed for on the appropriate F707B or Maintenance Procedure: SNOW or Page & Line…	
Name and Rank / Grade (block capitals)	Signature

UNIT	Date of Weigh	Job / Task Number

Authorised Aircraft Weigher	
Name and Rank / Grade (block capitals)	Signature

New basic figures have been entered / forwarded for inclusion on MOD Form 751 / WRAM	
Name and Rank / Grade (block capitals)	Signature

NEW BASIC WEIGHT OF AIRCRAFT = Lbs	

NEW BASIC MOMENT OF AIRCRAFT = Lbs/ins	

NEW BASIC C of G OF AIRCRAFT = ins	

NEW % MAC (if required)	

Latitude Correction Figure	

Weighing Position	Platform Ser No	Indicated Load	Platform Zero	Lat Correct Load	Calibration Correction	Actual Load
FIRST WEIGH						
Nose						
				SUB TOTAL	Wt1 =	
Main u/c Pt						
Main u/c Stbd						
				SUB TOTAL	Wt2 =	
O/rigger Pt						
O/rigger Stbd						
				SUB TOTAL	Wt3 =	
				A/C TOTAL	w =	
SECOND WEIGH						
Nose						
				SUB TOTAL	Wt1 =	
Main u/c Pt						
Main u/c Stbd						
				SUB TOTAL	Wt2 =	
O/rigger Pt						
O/rigger Stbd						
				SUB TOTAL	Wt3 =	
				A/C TOTAL	w =	
THIRD WEIGH						
Nose						
				SUB TOTAL	Wt1 =	
Main u/c Pt						
Main u/c Stbd						
				SUB TOTAL	Wt2 =	
O/rigger Pt						
O/rigger Stbd						
				SUB TOTAL	Wt3 =	
				A/C TOTAL	w =	

Median of 3 recorded aircraft Total weights = (w)	Lbs	w	
Nose weight from Median total weight figures	Lbs	Wt1	
Main u/c weight from Median total weight figures	Lbs	Wt2	
O/rigger weight from Median total weight figures	Lbs	Wt3	
Centre line of nose wheels to WRD	ins	r	
Centre line of Main u/c to WRD	ins	s	
Centre line of O/riggers to WRD	ins	t	
Nose u/c moment about WRD = Wt1 x r	Lbs/ins		
Main u/c moment about WRD = Wt2 x s	Lbs/ins		
O/rigger moment about WRD = Wt3 x t	Lbs/ins		
Total aircraft moment about WRD (WRDm)	Lbs/ins		
C of G from WRD = WRDm/w	ins	Xa	
Tail down angle (TDA) of Median total weight as a decimal		Degrees	
Vertical C of G position relating to Aircraft Mk and Configuration		Z	
C of G Coversion = Xa / cosTDA - (Z x tanTDA)			
Distance from WRD to C of G Datum	ins		
Converted C of G from C of G datum	ins	x	
As weighed moment = (x) x (w)	Lbs/ins	m	
Basic weight = (w) plus deficiencies, minus surpluses	Lbs	BW	
Basic moment = (m) plus deficiencies, minus surpluses	Lbs/ins	BM	
Basic Centre of Gravity from aircraft datum point = (BM) / (BW)	ins	X	
% Mean Aerodynamic Chord (%MAC) =		% MAC	